

Amendments To The Claims:

Please amend the claims as shown.

1 – 8 (canceled)

9. (new) A method for repairing an at least part-composite ceramic turbomachine component, comprising:

dissolving a joint of the component;  
mechanically processing the component to prepare a repair site;  
renovating the ceramic matrix of the component and restoring the joint; and  
filling the repair site with a single monobloc insert that is not a mat or a band and has a high strength where the edges of the repair and the edges of the insert converge flatly.

10. (new) The method as claimed in claim 1, wherein after the dissolving of the joint, the component is decoated before further processing.

11. (new) The method as claimed in claim 1, wherein mechanically processing the component consist of excavating the repair site.

12. (new) The method as claimed in claim 9, wherein the repair site is excavated by grinding.

13. (new) The method as claimed in claim 9, wherein the ceramic matrix is renovated by infiltration of the ceramic matrix.

14. (new) The method as claimed in claim 13, wherein the ceramic matrix is renovated by paste application.

15. (new) The method as claimed in claim 13, wherein weaving or recoating of the fibers is performed before the infiltration of the ceramic matrix.

16. (new) The method as claimed in claim 13, wherein weaving and recoating of the fibers is performed before the infiltration of the ceramic matrix.

17. (new) The method as claimed in claim 9, wherein the component is sintered before the joint is restored.

18. (new) The method as claimed in claim 9, wherein the component is coated before the step of restoring the joint.

19. (new) The method as claimed in claim 9, wherein surface protection is provided to the component after the step of restoring the joint.

20. (new) The method as claimed in claim 9, wherein the turbomachine component is completely formed from a composite ceramic material.

21. (new) A method for repairing a damaged or aged gas turbine component which is at least partially made of a composite ceramic material, comprising:

leaching out a matrix of the composite ceramic material or mechanically processing the component;

restoring or renovating the ceramic matrix of the component by infiltration; and  
sintering the component by overfiring the operating gas turbine.

22. (new) The method as claimed in claim 22, further comprising weaving or recoating of fibers of the component before restoring or renovating the ceramic matrix and after leaching out the matrix.

23. (new) A repaired gas turbine component which is at least partially made of a composite ceramic material, comprising:

a root portion;

a tip portion opposite the root portion;

an airfoil section arranged between the root and tip portions;

a repair section filled with a high strength single monobloc insert that is not a mat or a band where the edges of the repair section and the edges of the insert join smoothly and aerodynamically.

24. (new) The gas turbine component as claimed in claim 24, wherein the ceramic matrix of the component is restored or renovating by infiltration.

25. (new) The gas turbine component as claimed in claim 24, wherein the component is sintered by overfiring the operating gas turbine.